

WHAT IS CLAIMED IS:

1. A rotary turbojet blade that is to be subjected to a longitudinal gas flow, said blade comprising a plurality of blade sections extending along a line of the centers of gravity of said blade sections between a base and a tip of said blade, said blade being defined longitudinally between a leading edge and a trailing edge, said blade presenting along a radial axis of said turbojet a bottom portion, an intermediate portion, and a top portion, said bottom portion extending radially from said blade base to a bottom limit of said intermediate portion, and said top portion extending radially from a top limit of said intermediate portion to said blade tip, wherein said bottom portion presents a longitudinal angle of inclination for a leading edge line, said intermediate portion presents a backward longitudinal angle of inclination for said leading edge line, and said top portion presents a backward longitudinal angle of inclination for said leading edge line and a tangential angle of inclination for said line of the centers of gravity of the blade sections in a direction opposite to the direction of rotation of said blade.
2. A blade according to claim 1, wherein said bottom limit of the intermediate portion of the blade lies in the range 40% to 75% of the radial height of said blade between its base and its tip.
3. A blade according to claim 1, wherein the longitudinal angle of inclination of the leading edge line of said bottom portion lies in the range -5° to 15° relative to said radial axis of the turbojet.
4. A blade according to claim 1, wherein the backward longitudinal angle of inclination of the leading edge line of the intermediate portion lies in the range 5° to 20° relative to said radial axis of the turbojet.

5. A blade according to claim 1, wherein the backward longitudinal angle of inclination of the leading edge line of said top portion lies in the range 20° to 50° ,
5 and the tangential angle of inclination of the line of the centers of gravity of the blade sections of said top portion lies in the range 20° to 50° relative to said radial axis of the turbojet.
- 10 6. A blade according to claim 1, wherein said bottom portion further presents a tangential angle of inclination for the line of the centers of gravity of the blade sections lying in the range -5° to 15° relative to
15 said radial axis of the turbojet.
7. A blade according to claim 1, wherein said intermediate portion further presents a tangential angle of inclination for the line of the centers of gravity of the blade sections lying in the range -5° to 15° relative
20 to said radial axis of the turbojet.
8. A blade according to claim 1, wherein said top portion further comprises a top zone extending radially to said blade tip in which the leading edge line presents a
25 forward longitudinal angle of inclination.
9. A rotary turbojet machine for passing a flow of gas, including a plurality of blades according to claim 1.
- 30 10. A machine according to claim 9, constituting a turbojet fan.
11. A machine according to claim 9, constituting a turbojet compressor.